

Amendments To The Claims:

Please amend the claims as shown.

1 – 9 (canceled)

10. (new) A combustion chamber for a gas turbine, comprising:  
a combustion chamber wall;  
a liner formed from a plurality of heat shields on an inside of the combustion chamber wall;  
an inner space formed between the heat shield elements and the combustion chamber wall and exposed to a cooling medium; and  
a flow element arranged in the inner space for selective adjustment of a cooling medium stream.

11. (new) The combustion chamber as claimed in claim 10, wherein a flow channel for cooling medium is formed by the flow element causing a flow velocity of the cooling medium stream to be increased compared with the flow velocity upstream of the flow element.

12. (new) The combustion chamber as claimed in claim 10, wherein a heat shield element is assigned a respective flow element for the purpose of cooling a thermally heavily loaded wall section of the heat shield element.

13. (new) The combustion chamber as claimed in claim 12, wherein the heat shield element is a single-shell hollow vessel with a cavity that the flow element is disposed.

14. (new) The combustion chamber as claimed in claim 12, wherein the heat shield element has a surface region with a surface contour curved along a longitudinal axis and a transverse axis.

15. (new) The combustion chamber as claimed in claim 10, wherein the flow element is mounted with a positive fit on the combustion chamber wall.

16. (new) The combustion chamber as claimed in claims 10, wherein the flow element is detachably connected to the combustion chamber wall.
17. (new) The combustion chamber as claimed in claim 10, further comprising a flow element made of metal, in particular a metal sheet or a metal shaped part.
18. (new) The combustion chamber as claimed in claim 17, wherein the flow element is made of a metal sheet or a metal shaped part.
19. (new) A gas turbine having a combustion chamber, comprising:  
a combustion chamber wall;  
a liner formed from a plurality of heat shields on an inside of the combustion chamber wall;  
an inner space formed between the heat shield elements and the combustion chamber wall and exposed to a cooling medium; and  
a flow element arranged in the inner space for selective adjustment of a cooling medium stream.
20. (new) A flow element arranged in a flow channel between a combustion chamber wall and a heat shield element in a combustion chamber of a gas turbine, comprising:  
a surface of the flow element located near a cold side of the heat shield such that the flow channel becomes more narrow;  
a surface contour of the surface adapted to approximately match a surface contour of the cold side of the heat shield element.
21. (new) The flow element as claimed in claim 20, wherein a cooling medium flowing in the flow channel is caused to accelerate as the cooling medium flows by the surface.
22. (new) The flow element as claimed in claim 20, wherein the flow element is approximately rectangular in shape and the surface forms the longer side of the rectangle.

23. (new) The flow element as claimed in claim 20, wherein a heat shield element is assigned a respective flow element for the purpose of cooling a thermally heavily loaded wall section of the heat shield element.

24. (new) The flow element as claimed in claim 20, wherein the heat shield element is a single-shell hollow vessel with a cavity that the flow element is disposed.

25. (new) The flow element as claimed in claim 20, wherein the flow element is approximately triangular in shape and the surface forms the longer side of the triangle.

26. (new) The flow element as claimed in claim 20, wherein the surface is approximately parallel to the cold surface of the heat shield element.